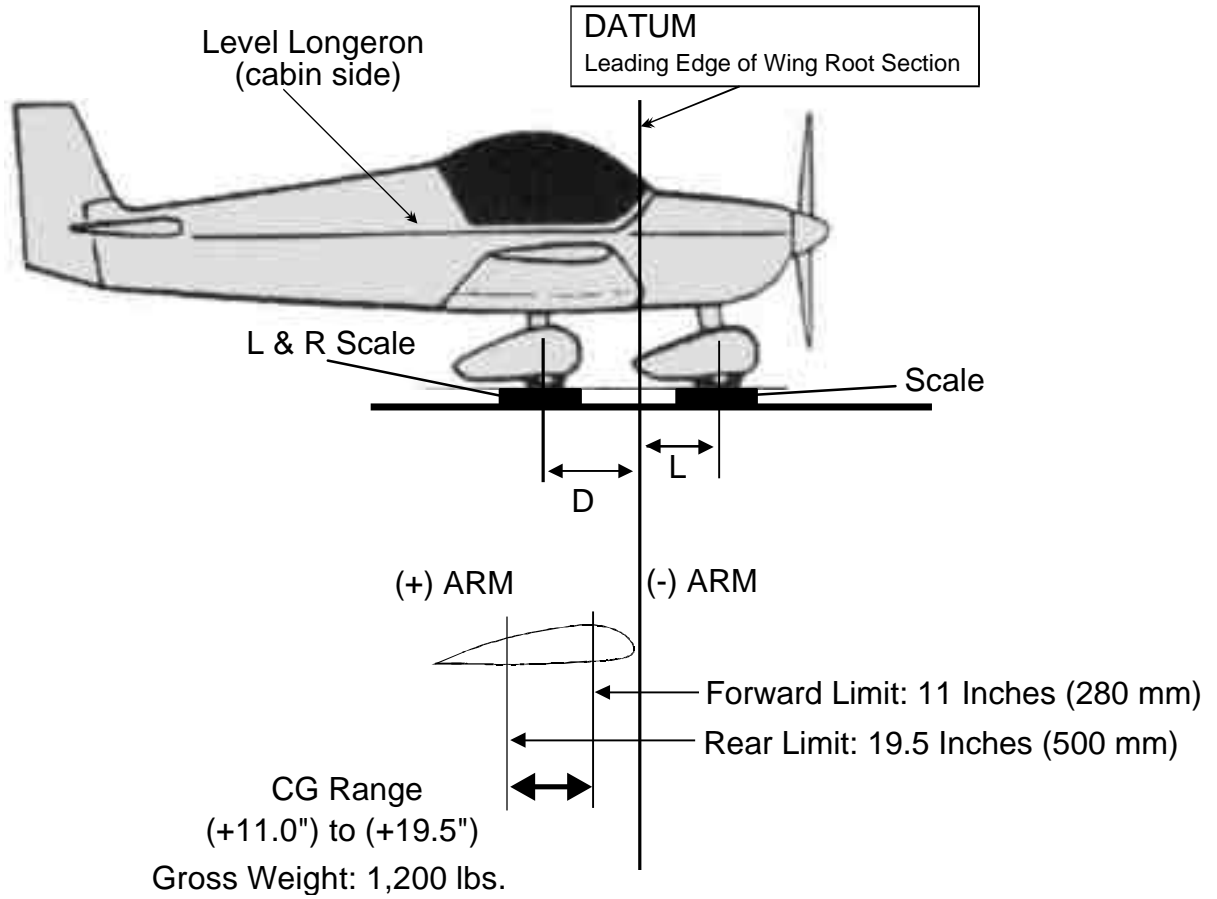




WEIGHT and BALANCE REPORT

ZODIAC CH 601 HDS – Tricycle Gear

Registration:
Serial No:



Forward Limit:	
Rear Limit:	
C.G. Range:	
Gross Weight	

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Weight & Balance Terminology:

DATUM: (Reference Datum) The imaginary vertical line from which all horizontal measurements are taken for balance purposes with the aircraft in level flight attitude. The datum is the leading edge (L.E.) of the wing (root) center section. All moment arms and the location of the permissible C.G. range must be taken with reference to the datum.

ARM: The horizontal distance from the reference datum line to the center of gravity of the item. The algebraic sign is plus (+) if measured aft of the datum, and minus (-) if measured forward of the datum.

MOMENT: The product of the weight of an item multiplied (x) by its arm.

CENTER OF GRAVITY: (C.G.) The point about which the aircraft would balance if it were possible to suspend it at that point.

CENTER OF GRAVITY LIMITS: The specified forward and aft points beyond which the C.G. must not be located during takeoff, flight, and landing.

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WEIGHT & BALANCE DATA OF EMPTY WEIGHT C.G.

Weighing Procedure

- All items which are not included as fixed equipment should be removed.
- Fuel tanks should be drained.
- Fill oil and cooling system to full capacity.
- With airplane on scales, block the main gear.
- Level the airplane: Level reference is the upper longeron at the cabin side.

Actual Measurements:

D= horiz. distance measured from the datum to the main wheels (26.5 inches)

L= horiz. distance measured from the datum to the nose wheel (19.5 inches)

WEIGHING POINT	Scale Reading (lbs)	- Tare (lbs)	= Weight (lbs)
RIGHT MAIN WHEEL			
LEFT MAIN WHEEL			
NOSE WHEEL			
TW =			

Computing the CG position for the aircraft empty weight:

WEIGHT	X ARM	= MOMENT
(Right Wheel)	(D)	
(Left Wheel)	(D)	
(Nose Wheel)	(L)	
TM =		

Divide the total moment TM by the total weight TW:

$$CG = TM / TW = \text{--- (CG Dist. Aft. Of Datum) --- Inches}$$

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WEIGHT & BALANCE EXTREME CONDITIONS

Forward Check:

ITEM	Weight (lbs)	X ARM (Inches)	= Moment
AIRCRAFT EMPTY	(TW Empty Aircraft)	(CG of Empty Aircraft)	(TM of Empty Aircraft)
PILOT		25.5"	
MAX FUEL (header tank)		-(9.5)"	
TOTAL	TW =		TM =

$$CG = TM / TW = \text{_____ Inches}$$

Rearward Check:

ITEM	Weight (lbs)	X ARM (Inches)	= Moment
AIRCRAFT EMPTY	(TW Empty Aircraft)	(CG of Empty Aircraft)	(TM of Empty Aircraft)
PILOT		25.5"	
PASSENGER		25.5"	
MIN FUEL (header tank)		-(9.5)"	
WING TANK (left & right)		23.75	
BAGGAGE (rear compartment)		55.1	
TOTAL	TW =		TM =

$$CG = TM / TW = \text{_____ Inches}$$

LIMITS:

Forward = 11"
 Rear = 19½"
 Max Weight = 1,058 (for 601 UL)

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