CH 601 XL / CH 650 B
WEIGHT & BALANCE REPORT
Forward & Aft C.G. Check

**AIRCRAFT EMPTY CG**

<table>
<thead>
<tr>
<th>ITEM</th>
<th>WEIGHT (pounds)</th>
<th>ARM (mm.)</th>
<th>MOMENT</th>
</tr>
</thead>
<tbody>
<tr>
<td>RIGHT MAIN WHEEL</td>
<td>WR =</td>
<td>LR=</td>
<td>WR x LR</td>
</tr>
<tr>
<td>LEFT MAIN WHEEL</td>
<td>WL =</td>
<td>LL=</td>
<td></td>
</tr>
<tr>
<td>NOSE WHEEL</td>
<td>WN =</td>
<td>LN= -</td>
<td>-</td>
</tr>
<tr>
<td>COMPUTED CG EMPTY</td>
<td>Empty Weight:</td>
<td>CG=</td>
<td></td>
</tr>
</tbody>
</table>

**PILOT**
- 710 mm.

**PASSENGER**
- 710 mm.

**BAGGAGE**
- 1,600 mm.

**FUEL: L.E. WING TANKS**
- 180 mm.

**WING LOCKERS (OPTION)**
- 560 mm.

**TOTAL**
- WF=
- WR=

**CG Range:** From 270 mm. to 455 mm.

**Forward Check:**
Add to the empty weight & aircraft moment items located forward of the computed empty CG (ie. Fuel) plus the pilot.

**Center of Gravity (CG) = Total Moment / Total Weight**
**CH 601 XL / CH 650 B: Center of Gravity Limits**

Diagram is not to scale

Datum: Wing Leading Edge at STN: 1,790 mm.

Computed C.G. = \( x \) mm.

MAC = 60 in. = 1.52 m.